WHAT IS CLAIMED IS:

1. A blade structure in a gas turbine, comprising:

stationary blades arrayed in a circle on a casing;

moving blades arrayed in a circle on a rotor, wherein a clearance is provided between chips of the moving blades and the casing, wherein

a front-edge including angle at a chip portion of the stationary blade that is the stationary blade at the rear stage of the moving blade having the chip clearance is larger than a front-edge including angle at other portions than the chip portion of the stationary blade.

2. A blade structure in a gas turbine, comprising:

stationary blades arrayed in a circle on a casing;

moving blades arrayed in a circle on a rotor, wherein a clearance is provided between chips of the moving blades and the casing, wherein

an entrance metal angle at a chip portion of the stationary blade that is the stationary blade at the rear stage of the moving blade having the chip clearance is smaller than an entrance metal angle at other portions than the chip portion of the stationary blade.

3. A blade structure in a gas turbine, comprising:

stationary blades arrayed in a circle on a casing;

moving blades arrayed in a circle on a rotor, wherein a clearance is provided between chips of the moving blades and the casing, wherein

a front-edge including angle at a chip portion of the stationary blade that is the stationary blade at the rear stage of the moving blade having the chip clearance is larger than a front-edge including angle at other portions than the chip portion of the stationary blade, and also an entrance metal angle at a chip portion of the stationary blade is smaller than an entrance metal angle at other portions than the chip portion of the stationary blade.

4. A blade structure in a gas turbine, comprising:

stationary blades arrayed in a circle on a casing;

moving blades arrayed in a circle on a rotor, wherein seal-air flows from the rotor side at the upstream of the moving blades, wherein

a front-edge including angle at a hub portion of the stationary blade is larger than a front-edge including angle at other portions than the hub portion of the moving blade.

5. A blade structure in a gas turbine, comprising:

stationary blades arrayed in a circle on a casing;

moving blades arrayed in a circle on a rotor, wherein seal-air flows from the rotor side at the upstream of the moving blades, wherein

an entrance metal angle at a hub portion of the stationary blade is smaller than an entrance metal angle at other portions than the hub portion of the moving blade.

6. A blade structure in a gas turbine , comprising:

stationary blades arrayed in a circle on a casing;

moving blades arrayed in a circle on a rotor, wherein seal-air flows from the rotor side at the upstream of the moving blades, wherein

a front-edge including angle at a hub portion of the stationary blade is larger than a front-edge including angle at other portions than the hub portion of the moving blade, and also an entrance metal angle at a hub portion of the stationary blade is smaller than an entrance metal angle at other portions than the hub portion of the moving blade.

7. A blade structure in a gas turbine , comprising:

stationary blades arrayed in a circle on a casing;

moving blades arrayed in a circle on a rotor, wherein a clearance is provided between chips of the moving blades

and the casing, wherein

a chord length at a chip portion of the moving blade having the chip clearance is larger than a minimum chord length at other portions than the chip portion of the moving blade.

8. The blade structure in a gas turbine according to claim 7, wherein the chip portion of the stationary blade is provided with an escape section for avoiding an interference with the chip portion of the moving blade.

9. The blade structure in a gas turbine according to claim 8, wherein the escape section of the chip portion of the stationary blade is arranged such that an entrance metal angle at a chip portion of the stationary blade is smaller than an entrance metal angle at other portions than the chip portion of the stationary blade, and that the entrance metal angle at the chip portion of the stationary blade is directed toward the rear surface side of the stationary blade.